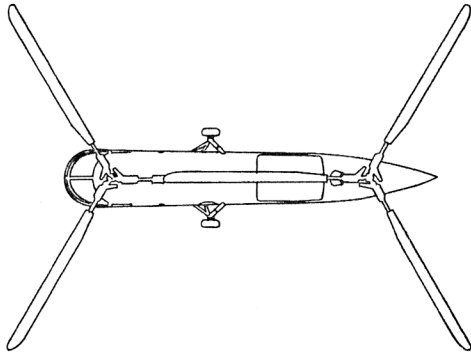
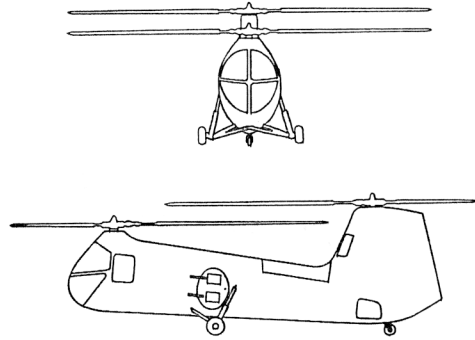


Characteristics Summary

HELICOPTER. H-25A



"ARMY MULE"



PIASECKI

Disc Area (projected) 1670 sq ft	Length (rotors operating) 56.9 ft
Rotor Dia 35.0 ft	(rotors static) 48.6 ft
Height 12.5 ft	(fuselage) 31.9 ft

AVAILABILITY			PROCUREMENT			
Number available			Number to be delivered in fiscal years			
ACTIVE	RESERVE	TOTAL				

STATUS

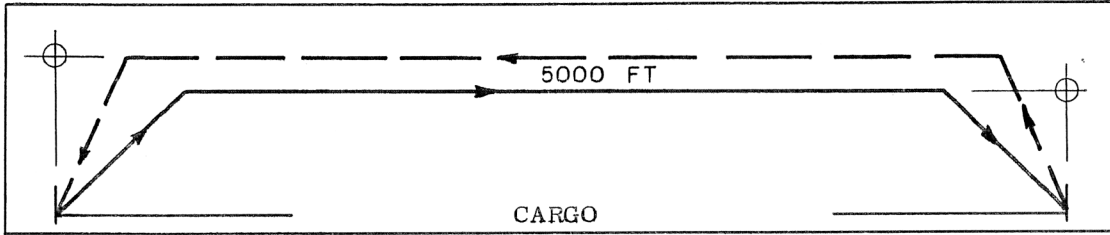
- Developed by the Navy and procured for the Dept of Army.
- Letter Contract: Apr 51
- First Flight: Aug 52
- First Acceptance: Oct 52
- Production Completed: Oct 53

Navy Equivalent: HUP-3

Mfr's Model: PV-18

POWER PLANT	FEATURES	GENERAL
(1) R-975-46 Continental ENGINE RATINGS BHP- RPM - ALT- MIN T.O: 475 - 2350 - SL - 30 Nor: 450 - 2350 - SL - Cont	Hydraulic Rescue Hoist External Cargo Sling Cargo Tie-down Provisions Dual Controls Hydra - Actuated Flight Controls Max Fuel Capacity: 150 gal	Crew (normal) 2 Pilot Co-pilot or Attendant Passengers 4 or Litters 3 Usable Cargo Space 160 cu ft Cargo Sling (ext) .. 1500 lb Limit Floor Load 60 lb/sq ft

Characteristics Summary Basic Mission H-25A



PERFORMANCE		
COMBAT RADIUS	COMBAT RANGE	S P E E D
28 naut. mi with 655 lb payload at 81 knots avg. in 0.8 hours.	69 naut. mi with 655 lb payload at 80 knots avg. in 0.9 hours	COMBAT 102 ^(b) knots at 5000 ft alt, nor power MAX 105 knots at zero ft alt, max power BASIC 102 ^(b) knots at 5000 ft alt, nor power
C L I M B	C E I L I N G	T A K E - O F F
910 fpm sea level, take-off weight normal power	11,700 ft 100 fpm, take-off weight normal power	ground run 0 ft — ft no assist assisted
1490 fpm sea level, combat weight Take-off power	13,000 ft 500 fpm, combat weight normal power	over 50 ft height 0 ft — ft no assist assisted
L O A D	W E I G H T S	H O V E R I N G C E I L .
Cargo: 655 lb Fuel: 39.2 gal protected 0 % droppable 0 % external 0 %	Empty..... 3930 lb Combat... 4631 lb Take - off 5400 lb limited by vertical rate of climb at 370 ft/min on a Navy Standard-day	9300 ft combat weight, T.O. power V E R T I C A L C L I M B 370 fpm sea level, take-off weight T.O. power

N O T E S

1. Performance Basis:
 (a) USAF Phase IV flight tests.
 (b) Limited by blade stall.

2. REVISION BASIS: To reflect latest performance and characteristics data.